

ABSTRACT

A gas turbine, particularly an aircraft engine includes at least one compressor. The gas turbine comprises at least one stator, at least one rotor and at least one generator (19) for generating electrical energy. An engine rotor has a rotor shaft (11) and rotor disks (12, 13, 14), which are driven by the rotor shaft (11) and which have rotating rotor blades (15). A stator has a housing (17) and fixed guide vanes (18). A generator (19) has at least one stator (21) and at least one rotor (20). The electrical energy generated by the generator (19) preferably serves to operate at least one attachment or one auxiliary unit of the gas turbine. According to the invention, the generator (19) is integrated in the interior of the gas turbine in such a manner that each rotor (20) of the generator is allocated to the compressor rotor and the stator (21) of the generator is allocated to the compressor stator, whereby kinetic energy of the rotor is convertible into electrical energy by the generator (19).